

MARS

MARS PATHFINDER AIRBAG IMPACT ATTENUATION SYSTEM*

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Abstract

The Mars Pathfinder spacecraft, scheduled for launch in November 1996, is designed to validate a low cost Entry, Descent, and Landing system and to perform scientific surface operations. The Jet Propulsion Laboratory and Sandia National Laboratories teamed to design, fabricate, test and validate a prototype 0.38 scale model of an airbag impact attenuation system.

A computer code was developed to predict the performance of the airbag system. A test program in Sandia's High Altitude Chamber was performed to validate the code and demonstrate the feasibility of the airbag concept and design. In addition, freefall tests were performed at representative velocities to demonstrate the structural integrity of the airbag system design. The feasibility program demonstrated that the airbag impact attenuation design will protect the lander upon impact with the Martian surface.

Nomenclature

a	linear acceleration of payload (m/s ²)
a*	speed of sound in gas (m/s)
A	Area (m ²)
A _{pp}	Area of bottom bag footprint (m ²)
g _{Mars}	acceleration of gravity on Mars (m/s ²)
k	discharge coefficient for an orifice
m	mass of the payload (kg)
M	Mach number
P	pressure (Pascals, newtons/m ²)

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R _{gas}	gas constant (Joules/kg K)
T	gas temperature (K)
t	time (seconds)
u	velocity of gas (333/s)
u	velocity of the payload (m/s)
v	volume (m ³)
w	mass of gas contained in airbag (kg)
x	distance (m)
γ	ratio of specific heats, (C _p /C _v), 1.4 for N ₂
ρ	density of gas (kg/m ³)

Introduction

The Mars Pathfinder spacecraft, scheduled to be launched in November 1996, will be the first lander launched to Mars since the Viking missions in the early 1970's. This mission, the first of a series of low cost Discovery class missions is a "pathfinder" for NASA's engineering design methodology of "faster, better, cheaper". The mission was designed to demonstrate and validate a low cost Entry, Descent, and Landing (EDL) system and to perform state-of-the-art scientific surface operations. The Entry and Descent designs are based on the Viking lander heat shield and parachute designs, respectively. The Landing portion of the spacecraft consists of a bridle mounted retro-rocket system and an airbag impact attenuation system.

The Jet Propulsion Laboratory teamed with Sandia National Laboratories to design, fabricate, test, and validate a 0.38 scale prototype airbag impact attenuation system as a proof of concept for the Pathfinder spacecraft. This paper will describe the airbag design, the computer model used for predicting airbag performance, and the tests conducted at Sandia in the High Altitude Chamber and the Coyote Canyon Test Facility. The airbags and tests were designed to validate the computer model as well as to demonstrate the structural integrity and dynamic characteristics of the system.

Concepts

Several concepts were examined for the descent and landing phase of the Pathfinder mission. A

system, modeled on the Viking system, with rockets actively controlling the vertical and horizontal components of velocity was considered. This system would have required complex componentry which became impractical given the schedule and budget available.

The driving design parameters were that the lander be able to deploy a rover onto the surface and also take a panoramic photo of the surface. Equally as important was that the lander not experience more than 50 g's deceleration on impact. The decision was made to use an inflatable impact attenuation system ~~was designed to~~ tolerate both vertical and horizontal velocities on impact. Since the horizontal velocity component at impact was estimated to be quite severe, a system with omnidirectional impact capability was required.

Spherical airbags were considered. Their downfall was that the payload would be completely surrounded by the inflatable membrane and the task of deploying the rover and instruments became very complex. Systems where multiple spherical airbags were simply clustered around the payload proved to be very inefficient. Non-spherical airbags of various geometric shapes were also considered and ruled out since they were structurally inefficient. The configuration finally selected consists of a tetrahedral lander with an airbag on each of its four sides. Each airbag is made up of spherical lobes merged together to form a single larger volume. The airbags are tethered to the lander via structural tendons integrated into each airbag. The system is inflated with hot gas generators approximately 3-8 seconds prior to impact.

The tetrahedral lander with four individual airbags allows access to the lander when the airbags deflate after the impact event. The rover and the scientific instrumentation are subsequently deployed by folding out three of the four tetrahedron panels, like a flower blossoming. This "self righting" method of opening meets the requirement to control the orientation of the lander as it comes to rest after landing. With the basic concept selected, the task of designing the airbags became a function of the actual impact conditions.

Impact Environment

The surface ambient conditions for the landing sites under consideration include a nighttime

temperature of approximately -75°C and a pressure of approximately 1kPa (about 1/100 that of Earth). Due to this rarefied atmosphere a practically sized parachute can only decelerate the spacecraft to a terminal velocity of about 35 m/s. Previously collected data indicate that 90th percentile surface winds are approximately 35 m/s. Those winds, combined with the vertical velocity, create a maximum impact velocity of 50 m/s at an angle of 45° . Rocks at the landing sites are predicted to be about 0.5 meters high.

Airbag Design

The original airbag design concept was derived from the "airbag impact attenuate?" work performed at JPL in the mid 1960's [1]. The fundamental concept, which is that of a "lumpy" sphere, implies a high modulus inflatable membrane with preloaded tendons. The tendons are used to transfer the pneumatic load from the membrane to the rigid body being decelerated.

Basic Sizing

The first generation Pathfinder airbag was sized to decelerate a 230 kg lander from 35 m/s to zero and limit the lander body deceleration to less than 50 g's. The first consideration in sizing the airbags was to determine the required deceleration stroke. Assuming a constant deceleration the required stroke is 1.25 meters. The mission requirement to survive impact on 0.5 meter rocks brings the minimum allowable airbag dimension to 1.75 meters,

Load transfer was maximized by minimizing relative movement between the airbag system and the lander. This was done by sizing and spacing the lobes of the airbags such that the center to center distance between any two adjacent lobes in the entire system was identical. Figure 1 shows the geometry for an airbag with three lobes, each with a radius of 1.2 meters and a lobe to lobe spacing of 1.92 meters. A single airbag is composed of three spherical lobes as seen from the outside. Internally, however, the airbag is one open volume with no internal membranes or septums. Figure 2 shows the geometry of the entire airbag system.

Once the basic configuration was selected, analysis was performed to characterize the performance. Internal vents were added to allow the

"bottom" bag to vent to each of the three top bags so that all of the airbag volume could be acted upon during impact. This resulted in three ducts emerging from the bottom bag, each connecting to one of the top airbags.

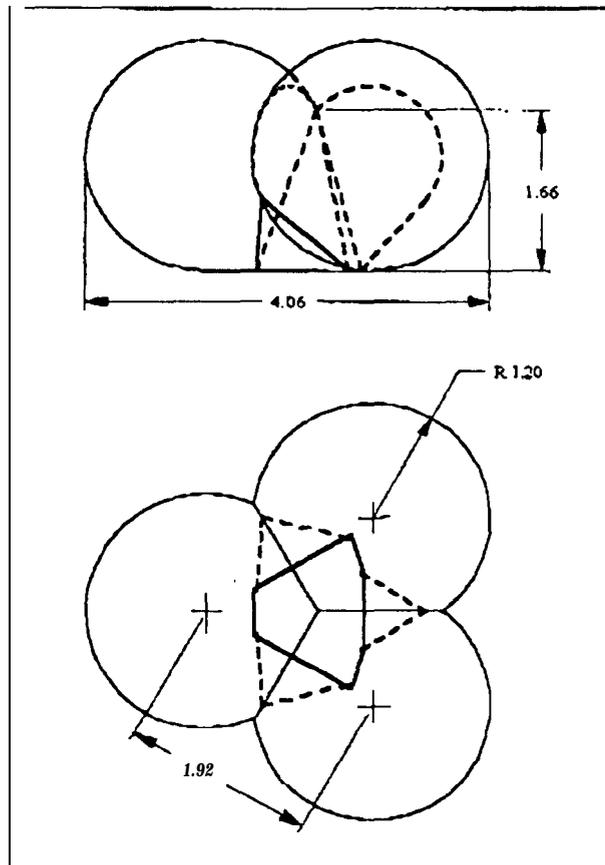


Figure 1- Three-lobe airbag design

Further analysis of the difference between the constant deceleration profile assumption made earlier and the more refined performance analysis showed that the 2.4m diameter dimension of the airbag was not sufficient to provide an adequate clearance margin for a nominal impact. This led to the seared design enhancement, external venting. Controlled venting of the gas from the bottom bag to the ambient atmosphere allowed a higher initial pressure to be used without exceeding the deceleration requirement. The original venting scheme called for the maximum pressure condition to exist prior to impact. On impact the external vents would be opened. The veins were sized so that the mass flow leaving the airbags matched the decrease in volume during an impact, thus yielding a nearly constant pressure.

An additional lightweight gas tight internal diaphragm was installed in the bottom airbag (see figure 6). The diaphragm is inflated into the bottom bag by gas from the three side bags during the first bounce, halts the external venting, and insures that the airbag system remains inflated for subsequent bounces.

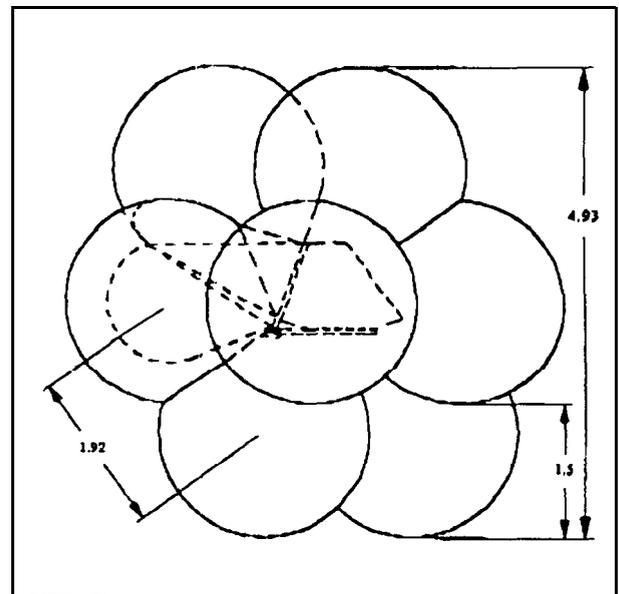


Figure 2 - Airbag system with lander.

Construction

Polyurethane coated 3.0 oz/yd² Kevlar fabric was selected for the proof of concept 0.38 scale prototype airbag. The lobes were created by a combination of fabric patterning and tensioning of the tendons anchored to the lander. The fabric patterns were joined using simple lap joint seams, formed by pressure fusion welding the urethane coating on the Kevlar fabric.

The airbag fabric was ^{nearly all} not directly attached to the lander in any location. Much of the pneumatic force was transferred from the membrane to the lander via the tendons, which are shown in Figures 1 and 2. Six tendons originate at each of six hardpoints on the lander, follow the external surface of the bag, and then pair up at a confluence point approximately 1/3 the way around the bag. Three tendons then follow the valley between lobes up to the cusp. At the cusp, the tendons pass through the fabric and once again divide into six different tendons which pass through the fabric again and

attach at each of the six hardpoints. ~~It is~~ the preload, in the tendons, generated by the inflation pressure which transfers much of the federation forces to the lander. Therefore, it was important to maximize the stiffness of the tendons, which was accomplished by using a high elastic modulus material. Kevlar.

Second Generation Airbag Design

Maturity of both the landing site location information and the entry and descent designs, allowed a relaxation of the impact requirements. Both the maximum vertical and horizontal velocities expected at impact were reduced to 20 m/s. This allowed a redesign of the airbags to make them smaller and lighter. The new design was based on the same design principal as the original three-lobe design. The new design consists of six spheres meshed together to form a single airbag. The internal venting scheme was preserved, however, the external venting was no longer required to provide the necessary margins. Figures 3,4, and 5 show the six-lobe configuration.

Computer Modeling of the Airbag Impact Attenuation System.

A computer model was developed at Sandia to model the pneumatic performance of the airbags and the rigid-body dynamics performance of the payload during ground impact. In Figure 6, it can be seen that the forces acting on the payload consist of the payload weight acting downward and the pressure force acting upward. Applying Newton's second law, the equation of motion for this system is obtained:

$$a = g_{Mars} - \frac{A_{FP} P_o}{m} \left(\frac{F_{P_{bag-1}}}{P_o} \right) \quad (1)$$

The nitrogen gas that was contained in all of the bags was assumed to perform as a perfect gas. The mass of gas contained in each bag was calculated from the known gas pressure, temperature, and bag volume.

Because all of the top bags communicated pneumatically with the bottom bag, the gas contained within them was assumed to be at the same pressure and temperature before impact. Just before impact, external orifices could be opened in

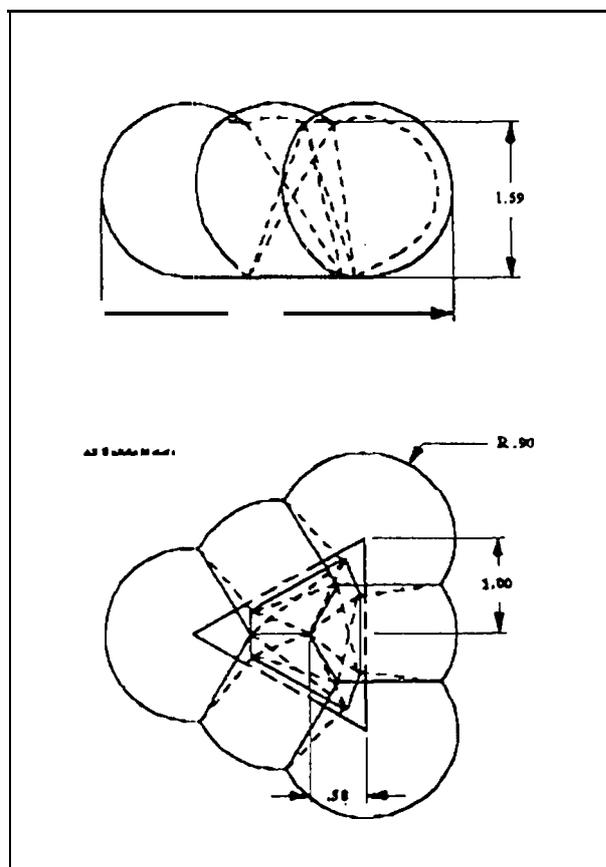


Figure 3- Six-lobe airbag.

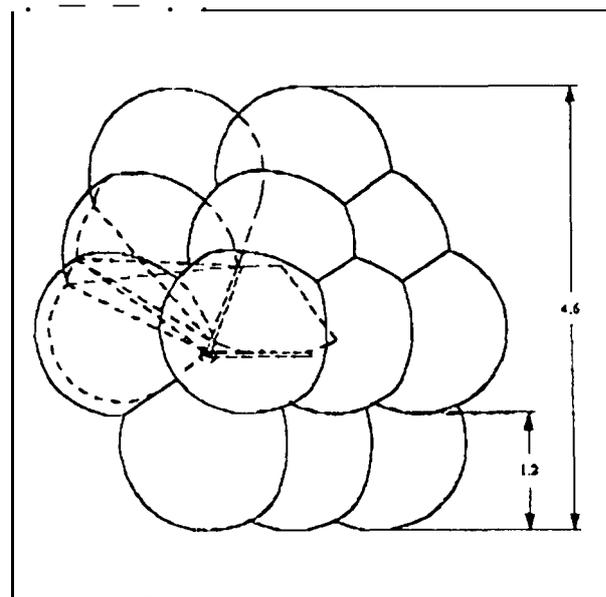


Figure 4- Six-lobe airbags mounted to lander.

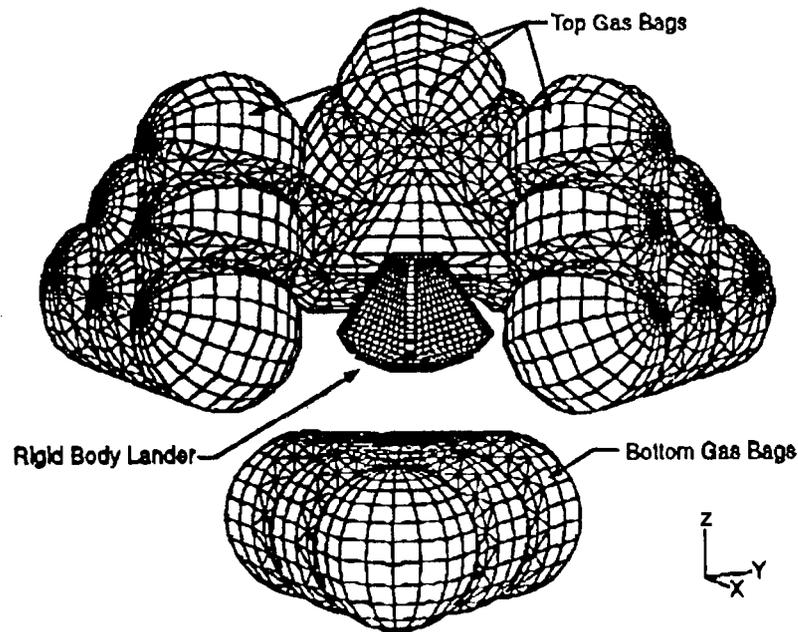


Figure 5- Exploded view of Dyna-3D finite element model, sk-lobe design
(Model created by Marc Collier and John McKinney of Rockwell Aerospace Space Systems Division)

the bottom bag and gas could start flowing out to the local environment from the bottom bag. At impact, the volume of the bottom bag began to decrease. If the external orifices were not too large, the gas pressure in the bottom bag would begin to increase forcing gas back into the top bags as well as to the outside. If the external orifices were too large, the pressure in the bottom bag would continue to decrease and the gas in the top bags would continue to flow into the bottom bag until the diaphragm expanded to fill the remaining volume of the bottom bag. At that time gas would cease flowing from the external orifices.

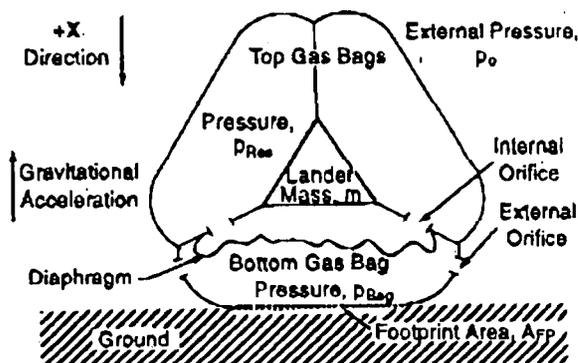


Figure 6- Impact Attenuation System Sketch

Gas flow through an orifice can be subsonic or sonic. For a diatomic gas, such as nitrogen, the ratio of the static pressure of the gas downstream to that upstream is greater than 0.5283, the flow is subsonic up to and through the smallest flow area, the orifice. If the ratio is smaller than 0.5283, the flow is subsonic as it approaches, but sonic in the orifice [2].

To slow the impacting payload, its kinetic energy must be transferred into the potential energy of the compressed gas. Thus, the airbags must operate with an internal pressure that is greater than the local atmospheric pressure to generate a form which, acting through the deflection of the bag, does work on the payload. A gasbag impact attenuation system intended for use on Mars can be tested in the earth's atmospheric pressure with the intended initial pressure differential from the bag interior to the atmosphere, but the external orifices will not perform as they would on Mars since the pressure ratio across them is significantly different.

The mass of gas discharged through the external orifices and the mass of gas remaining in the bag directly affect the rebound of the payload, so an atmospheric test on earth would not exhibit the same initial rebound speed as would actually occur on Mars.

The mass flow rate at the orifice throat is:

$$\frac{dW}{dt} = kA_{th}\rho_{th}u_{th} \quad (2)$$

The gas density at the throat of the orifice is:

$$\rho_{th} = \frac{P_{th}}{(R_{GAS} T_{th})} \quad (3)$$

and the gas velocity at the throat in terms of sonic conditions is:

$$u_{th} = M_{th} a_{th}^* \quad (4)$$

where:

$$a_{th}^* = \sqrt{\gamma R_{GAS} T_{th}} \quad (5)$$

When equations 3-5 are factored into equation 2, the mass flow rate is expressed in terms of the pressure and temperature in the orifice throat.

$$\frac{dW}{dt} = kA_{th}P_{th}M_{th}\sqrt{\frac{\gamma}{R_{GAS} T_{th}}} \quad (6)$$

Since the average gas flow within a bag is nearly zero, the static pressure and the stagnation pressure in the bag are essentially equal. Hence from reference 4 the static pressure as a function of the pressure upstream of the orifice is:

$$\frac{P_{th}}{P_u} = \left(1 + \frac{(\gamma-1)M_{th}^2}{2}\right)^{\frac{\gamma}{1-\gamma}} \quad (7)$$

When equation 7 is solved for M and substituted along with the adiabatic equation into equation 6, the equation for the mass flow rate through a subsonic orifice is obtained

$$\frac{dW}{dt} = kA_{th}P_d \left(\frac{1}{R_{GAS} T_d}\right)^{0.5} \left(\frac{2\gamma}{\gamma-1} \left(\frac{P_d}{P_u}\right)^{\frac{\gamma-1}{\gamma}}\right)^{0.5} \left(\left(\frac{P_u}{P_d}\right)^{\frac{\gamma-1}{\gamma}}\right)^{0.5} \quad (8)$$

For sonic flow in the orifice, $M=1.0$, and equation 7 reduces to:

$$\frac{P_{th}}{P_u} = \left(\frac{\gamma+1}{2}\right)^{\frac{\gamma}{1-\gamma}} \quad (9)$$

Substituting the adiabatic equation and equation 9 into equation 6 and rearranging produces the equation for the mass flow through a sonic orifice.

$$\frac{dW}{dt} = kA_{th}P_d \left(\frac{1}{R_{GAS} T_d}\right)^{0.5} \left(\gamma \left(\frac{2}{\gamma+1}\right)^{\frac{\gamma+1}{\gamma-1}}\right)^{0.5} \left(\left(\frac{P_u}{P_d}\right)^{\frac{\gamma+1}{\gamma}}\right)^{0.5} \quad (10)$$

Discharge Coefficient

Both equations 8 and 10 depend upon the value of the orifice discharge coefficient, k. Experiments were reported in reference 4 in which the pressure differential across a sharp-edged orifice was varied [0 obtain subcritical (subsonic) and critical (sonic) flow conditions, Experimental values of k measured as a function of the ratio of downstream pressure for the range from 0.0 to 1.0 were included in the model.

Each top bag can exchange gas only with the bottom bag through its internal orifices. The mass flow rate, dw/dt , is determined by equation 8 or 10 depending on whether the orifice flow is subsonic or sonic. If the pressure differential causes flow from the bottom bag into the three top bags (reservoir), that flow rate is considered to be negative and the mass of gas in the reservoir increases.

The bottom bag can exchange gas with the three side bags through the internal orifices and also with the outside through the external orifices. Gas flow out of the bottom bag whether through the internal or external orifices is considered to be negative. Thus, flow into the side bags and to the atmosphere will decrease the mass of gas contained in the bottom bag.

Change in Bottom Bag Area and Volume

AK 6095?
The gas bags proposed for the lander were constructed of fabric and remained with external and internal ~~teeth~~ so that each appeared to consist of three or more spherical bags merged together.

TENNONS
As shown in equation 1, the force that acts to decelerate the payload depends directly upon the area of the footprint, A_{FF} , of the bag with the ground.

The internal pressure in the bottom bag which is being **crushed** also depends directly **upon** the change in the bag's volume. **Note** that an implicit assumption made **in** developing the **area** and volume **equations** for **each bag** design **was** that the parts of the **bottom** bag not in contact with the ground did not move nor flex relative to the lander. The **area** of the footprint as a function of the **bag crushup** was calculated for each **bag configuration** based **strictly** upon the bag geometry and the area of the **bag** that would **be** intersected by **the plane** of the ground during **crushup**.

Bounce Equations

Sins the gas remaining in the bags after the **payload has been successfully stopped** is **still** at a pressure greater than the local atmospheric pressure, the payload experiences some **rebound (bounce)**. **The same governing equation** which calculates the **transfer of the** payload's kinetic energy **into** potential energy stored in the **compression** of the gas also **calculates** the **reverse transfer of** energy from the gas to the **payload**. The maximum velocity for a rebound is **achieved** at the **instant** the gas bag **ceases** to push on the payload, This **occurs** when **the** gasbag has reinflated to its original shape or when the pressure in the bag **reaches** the **local** atmospheric pressure. **The height of the bounce**, assuming no aerodynamic **drag**, is found by simply equating the payload's kinetic energy at the start of the rebound with its **potential** energy at the top of the **rebound**.

Modeling

The **feasibility** of using gas bags to cushion the **lander** payload during ground impact on Mars had to be proven through testing on **Earth**. A modeling study was made to determine what test **conditions** were **required** to obtain **meaningful data**.

The Buckingham Pi Theorem [6] states that the number of nondimensional numbers required to define a problem is **equal** to the total number of variables minus the **total** number of dimensions **involved**. There are **19** variables and four dimensions, **mass, length, time and temperature** in **this** problem, Thus, **15** nondimensional number are needed.

These non dimensional number **can** be obtained in many **ways** that **often** seem **somewhat arbitrary**. **Once generated**, they can be modified by multiplication or division with any **combination** of

the other nondimensional numbers that are involved. In this ease, the more **complicated non-dimensional** numbers were chosen from the **coefficients** in the nondimensional governing differential equations. Note that the product of the gas instant and the temperature was considered as one variable **since** they were the only ones in this problem **that constrained** the dimension of **temperature**. Doing this did **not affect** the quantity of nondimensional numbers **required**, because **both** the number of **variables** and the number of dimensions were **decrease** by one. Fifteen non-dimensional number were **obtained**.

Let the system being tested on **Earth** be the "model", denoted by the **subscript, M**, and the system that is to **impact** on another planet be the "prototype", denoted by the **subscript, P**. For the results of a model test to be **very meaningful** all of the model nondimensional numbers must equal their **counterparts** for the prototype. When this **occurs** the modeling is called "undistorted."

Assume that the geometric scaling factor between the **model** and the prototype is:

$$\lambda = \frac{(X_o)_M}{(X_o)_P}$$

and that the pressure **ratio** between the **model** and the **prototype** is:

$$n = \frac{P_M}{P_P}$$

Then equating the nondimensional numbers between the **model** and the prototype produces the following **relationships** between the **variables** of the model and the prototype:

$$\begin{aligned} X_M &= \lambda \cdot X_P & A_M &= \lambda^2 \cdot A_P \\ V_M &= \lambda^3 \cdot V_P & m_M &= n \cdot \lambda^3 \cdot m_P \\ W_M &= n \cdot \lambda^3 \cdot W_P & \gamma_M &= \gamma_P \\ (R_{GAS T})_M &= (R_{GAS T})_P & k_M &= k_P \\ P_M &= n \cdot P_P & t_M &= \lambda \cdot t_P \\ U_M &= U_P & a_M &= \left(\frac{1}{\lambda}\right) \cdot a_P \end{aligned}$$

The last equation shows the relationship **that** must be **preserved** between all of the model and prototype **accelerations**. Hence, for the modeling to **be undistorted**:

$$g_M = \left(\frac{11}{\lambda} \right) g_P$$

The gravitational acceleration for the model is that of Earth, and for the prototype, that of Mars where the full-scale lander will impact. Therefore, the ratio of the gravitational acceleration between Earth and Mars, $\lambda = 0.38$, prescribes the model scale for undistorted model testing in full Earth gravity.

The feature, n , was added to the modeling equations to free the model mass from having to be met by λ^3 times the prototype mass. This proved useful during the design of the High Altitude Chamber impact apparatus and in the development of the test program parameters.

These modeling relationships also prescribe how the measurements made during model impact tests can be used to predict the prototype impact on another planet during model impact. Prototype velocities are equal to model velocities. Prototype accelerations are λ times model accelerations. If the

same gas at the same initial temperature is used in the model and prototype, then temperature variations will be the same between the model and prototype impacts.

Prototype Airbag Test Program

High Altitude Chamber Test Series #L

A test series of the original design was performed in May 1993 in the Sandia High Altitude Chamber (HAC). The impact test apparatus shown in Figure 7 was designed, fabricated, and assembled in the HAC for these tests. Because accelerating the airbag system itself was impractical in the limited space of the HAC the "ground" was accelerated into the stationary, inverted lander/airbag system, which was mounted on a strut. The impact plate was accelerated into the stationary airbag system by bungee cords. After the pressure in the HAC was brought to test conditions the cords were stretched to equal tension and the impact plate was released through initiation of an explosive cutter. Just before impact the bungee cords went slack and the velocity

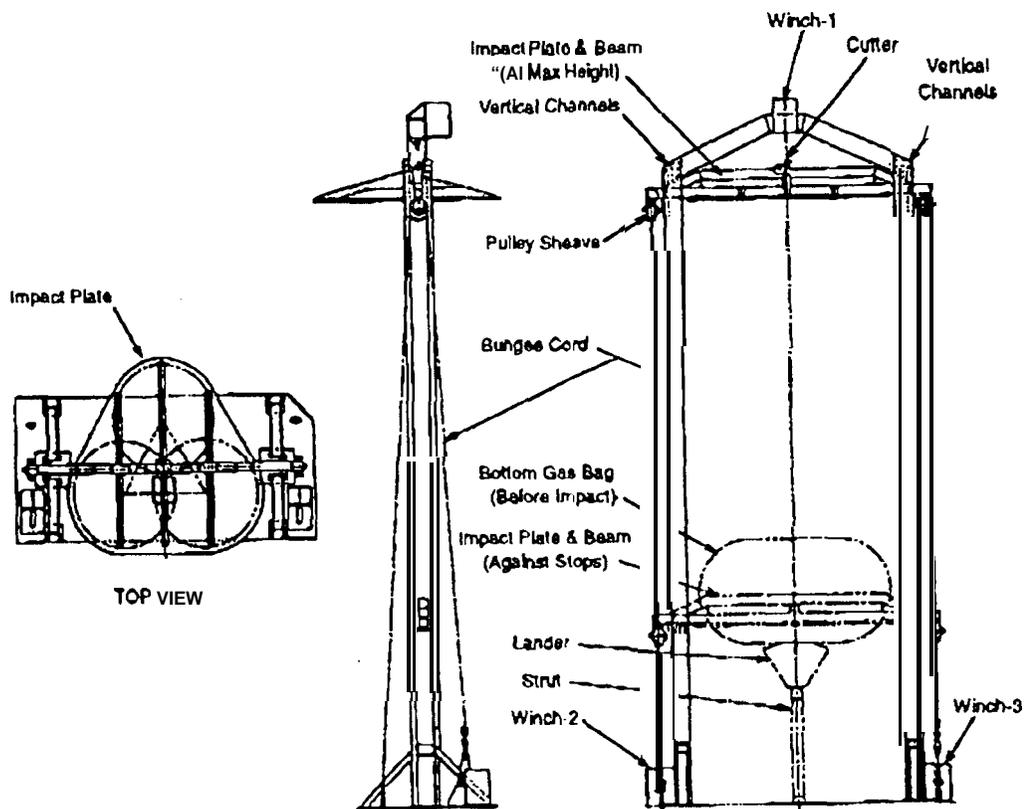


Figure 7- MESUR Impact Test Apparatus (top bags removed for clarity)

of the impact plate was measured by an optical velocity trap. Instrumentation on the impact plate and in each of the gas bags measured the deceleration and pressure-time history of the event.

The impact plate was designed for minimal mass, but still exceeded the scaled mass of the model by a factor of two. The scaling model therefore required that the pressures (both internal and external) be doubled to provide a suitable simulation of deceleration. The tests were performed at a pressure of 2.0 kPa (0.290 psi) external pressure, about twice that of the Martian atmosphere. The airbags were tested with initial internal pressures of 8.0 kPa (1.16 psi) and 12.0 kPa (1.74 psi),

Eighteen impact tests were performed with variations in impact velocity (maximum of 20 meters/second), external orifice areas, and internal orifice areas. The tests validated the mathematical computer model and demonstrated that the timing for external vent opening was critical. Those tests also showed that due to the location of the burst patches the opening of the vents was severely affected by the crush of the bags and was not repeatable. A modified code which tailored the external orifice area as a function of stroke as measured in the video documentation was prepared and used to model the ah-bag system.

Monte Carlo simulation - model!
exceed The results of these tests indicated that the performance of the airbag system was marginal for protecting the lander to less than the 50 g deceleration limit without the burst patches. A decision was made to add the retro-rocket system to the descent and landing system in lieu of the burst patches. The retro-rocket system decreases the vertical velocity component and allowed the ah-bags to be decreased in size.

Coyote Canyon Test Series

A test series was required to evaluate the structural integrity of the airbag impact attenuation system and demonstrate system feasibility. Free fall drops of the system were needed at representative velocities to meet this requirement.

The Coyote Canyon Test Facility at Sandia National Laboratories, which consists of a cable suspended across a valley between two mountain ridges, was chosen as the site for this test series. The cable facility provides a drop height of

approximately 183 meter (600 feet) at the highest point above the valley floor. In the initial test series the airbags were dropped vertically from the cable. A 147 kg (325 lb) mass was attached to the bottom of the airbag system to accelerate the system to a desired impact velocity. At impact the cable attaching the weight was severed so that the airbag system could rebound naturally.

The first test of the system at 12 m/s impact velocity was successful; however, the second 20 m/s impact test caused substantial damage to the airbag structure. It should be noted, however, that the lander was still protected during this impact and did not exceed the 50 g deceleration limit. The two primary causes for this damage was inadequate strain relief in the basic construction and faulty load paths at the tendon passthroughs. This problem was alleviated in the second prototype by creating a more direct tendon load path and using nylon fabric locally around the passthroughs. The flight design is avoiding rids problem completely by creating an enlarged inverted fabric boot to form the passthrough. The boot prevents any membrane loading in the vicinity of the hardpoint.

The second prototype bag was tested with both vertical and horizontal components of velocity. This test was conducted by accelerating the airbag system down an inclined wire on a trolley designed by JPL. Five tests were conducted at velocities up to 25.6 m/s total velocity. For the last test, a simulated Martian rocky surface was prepared. The airbags and lander survived all of these tests with the only damage being a four-inch tear in the fabric on the final test. Figure 8 shows an impact from these tests.

HAC Test Series #2

A second HAC test series was conducted in mid-September 1994. The purpose of this series was to demonstrate performance of the second generation airbag design under the same conditions as the original design and to validate the mathematical predictions used to size and design this second generation system. One of the major performance parameters investigated in this test series was the size of the internal orifices. The computer model predicts an optimal size which minimizes the federation and also the rebound. In addition, tests were conducted to demonstrate performance of the airbags when impact occurs on a side bag or on three bags at an apex.

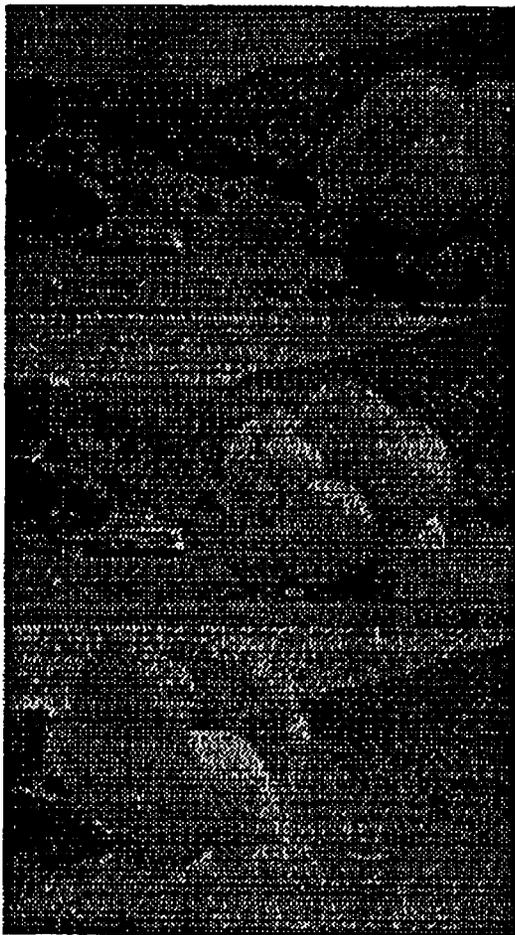


Figure 8- Canyon Test Airbag Impact

The same impact apparatus used for the first test series in the HAC was reused for this series of tests. In addition, a tripod stand was constructed to support the lander/airbag system with an apex pointed upward. This allowed investigation of an impact on the "apex" of the airbags where the footprint area impacting the ground is smallest.

The impact test results were then compared to the computer model. The results were not consistent with the computer model. However, examination of the test video showed that flexing of the airbags invalidated one of the code assumptions. A pneumatic volume module was added to the code to compensate for the main bag flexing below the original bottom plane of the lander. Also the footprint area and volume of the impact bag was calculated from the impact data and inserted into the computer model. These data more accurately modeled the area and volume of the system as it flexed away from the impact plate. Finally, the flow through the internal orifices of the second generation

airbags did not appear to be properly modeled by a sharp edged orifice. A constant flow coefficient of 0.98 was used to model this flow. This modified axle more accurately models the impact attenuation system results. Figures 9 and 10 illustrate the bottom bag pressure and deceleration test data and the code simulation curve.

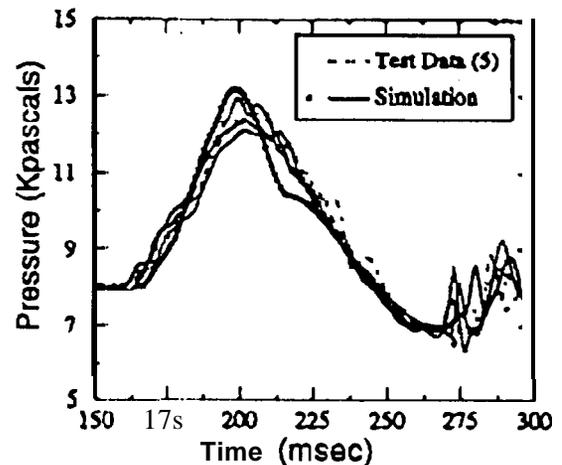


Figure 9- Pressure versus Time

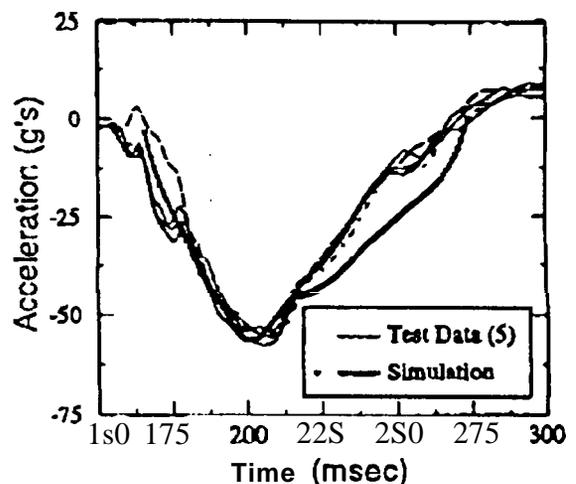


Figure 10 - Acceleration vs. Time

Flight System Testing

The test program described in this paper was considered proof of concept work to demonstrate the feasibility of using an airbag system for a planetary lander. In support of the flight article to be sent to Mars, JPL and ILC Dover Inc. have designed and are performing a much more extensive and detailed series of tests. These tests are all conducted with full scale airbags under Mars pressure conditions. The

tests include, but are not limited to; retraction testing, rapid inflation testing at Mars temperature and pressure, and full speed drop testing onto a 60° slanted platform with 0.5 meter high rocks. The results of these tests will validate and qualify the flight airbag and gas generator designs.

Conclusions and Recommendations

An impact attenuation system for the Mars Pathfinder spacecraft has been designed, a prototype 0.38 scale model fabricated, a computer model prepared, and tests conducted. This feasibility program has demonstrated that the Airbag Impact Attenuation design will protect the Mars Pathfinder lander upon impact with the Martian surface. A contract for the flight test bag fabrication and demonstration program has been issued as part of the preparations for the November 1996 launch.

The computer code developed and validated for performance of the airbag system provides a tool for investigating performance of the full-scale system in the Martian environment. Parametric studies can be performed to investigate variations in impact velocity, internal and external pressure, internal vent areas, and lander mass. Also, modifications to the airbag system geometry can be investigated by changing the area and volume inputs in the code. The computer code is currently limited, however, to investigation of vertical impacts on the bottom bag.

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